

1. (Currently Amended) An annular combustor for a gas turbine, comprising:
a wall, an inlet side, and an outlet side, the inlet side configured and arranged to accept burners opening on the inlet side, the combustor extending in an axial direction from the inlet side to the outlet side, cooled liner segments lining insides of the wall for protection from hot gases;

wherein the liner segments are subdivided in the axial direction into a plurality of parts arranged one behind the other; and

a plurality of segment carriers, the liner segments being fastened to the segment carriers, the segment carriers being subdivided in the axial direction into a plurality of separate, detachably connected parts.

2. (Previously Presented) The combustor as claimed in claim 1, wherein the liner segments are subdivided into two parts.

3. (Previously Presented) The combustor as claimed in claim 2, wherein the liner segments are subdivided where the flow velocity of the hot gases is low.

4. (Previously Presented) The combustor as claimed in claim 3, wherein the liner segments are subdivided in such a way that the lengths of individual segment parts in the axial direction are approximately the same.

5. (Cancelled)

6. (Previously Presented) The combustor as claimed in Claim 1, wherein the liner segments are convection-cooled.

7. (Previously Presented) The combustor as claimed in claim 6, wherein the subdivided liner segments are separately convection-cooled.

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